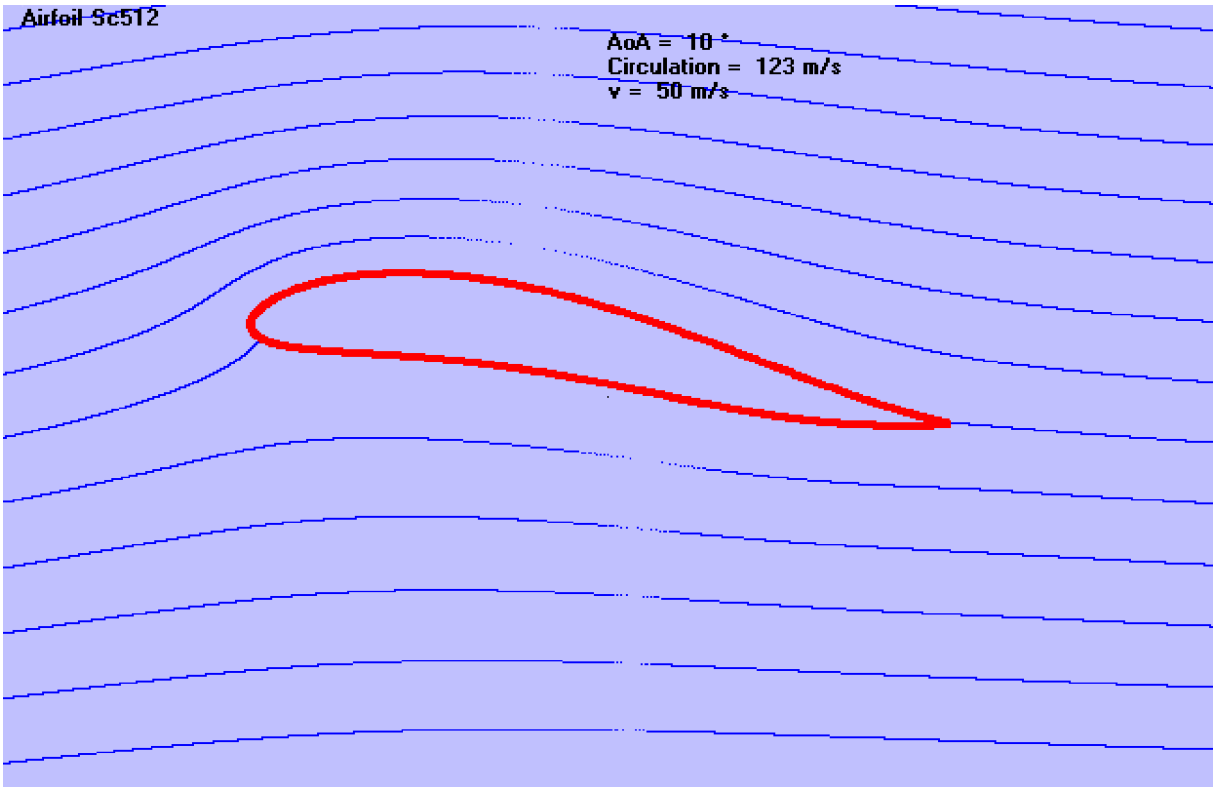
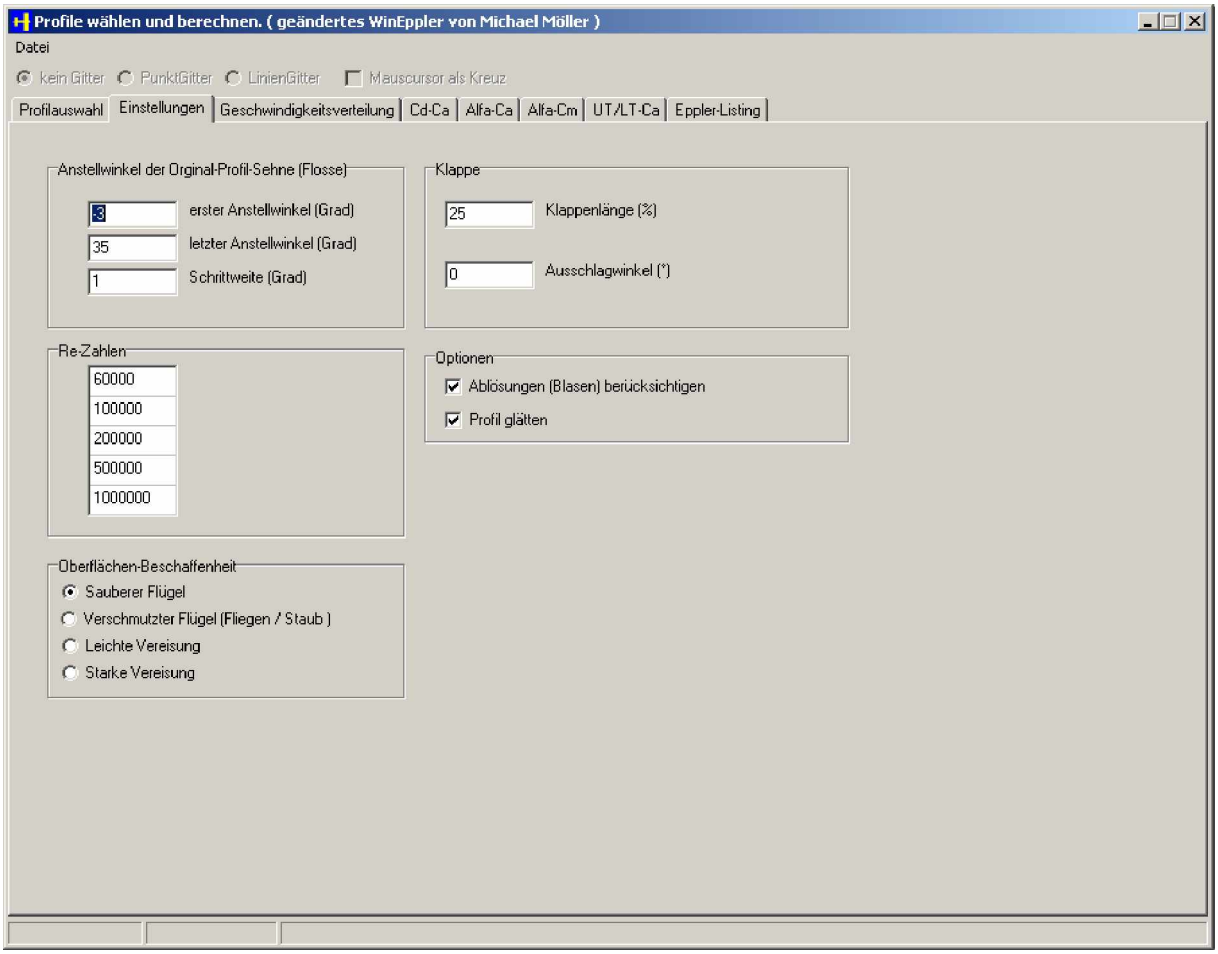
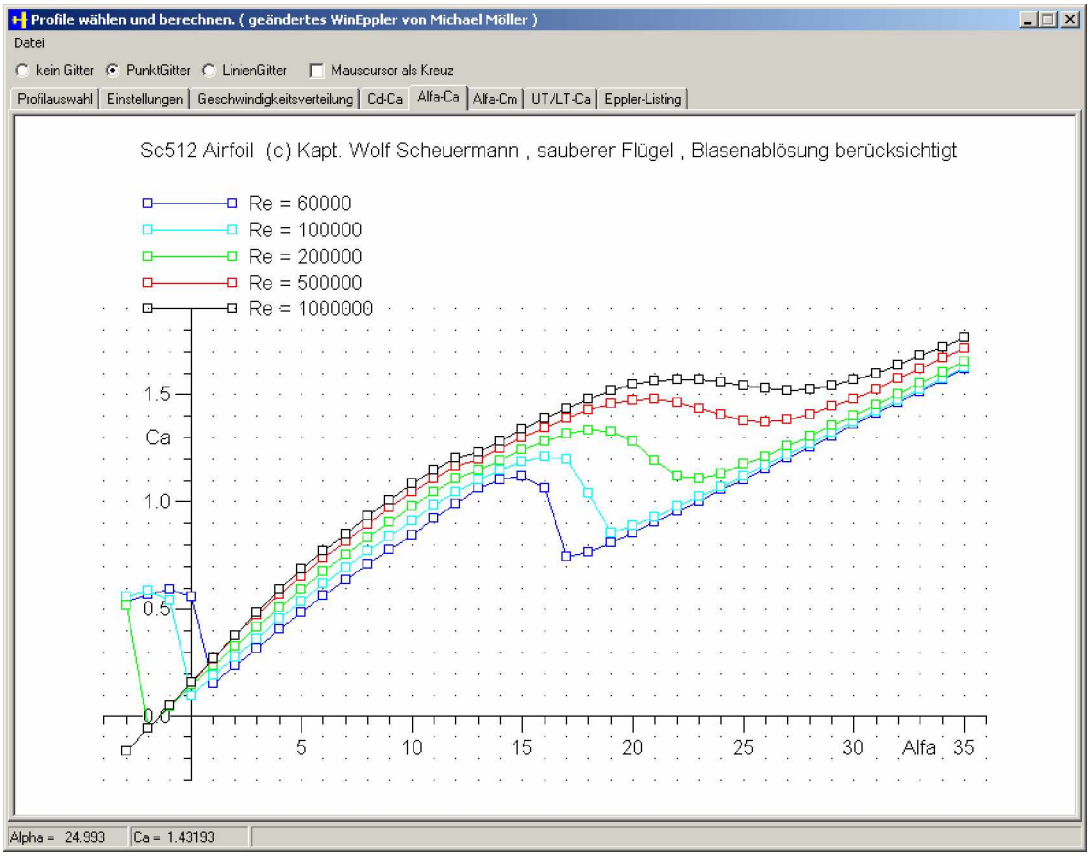
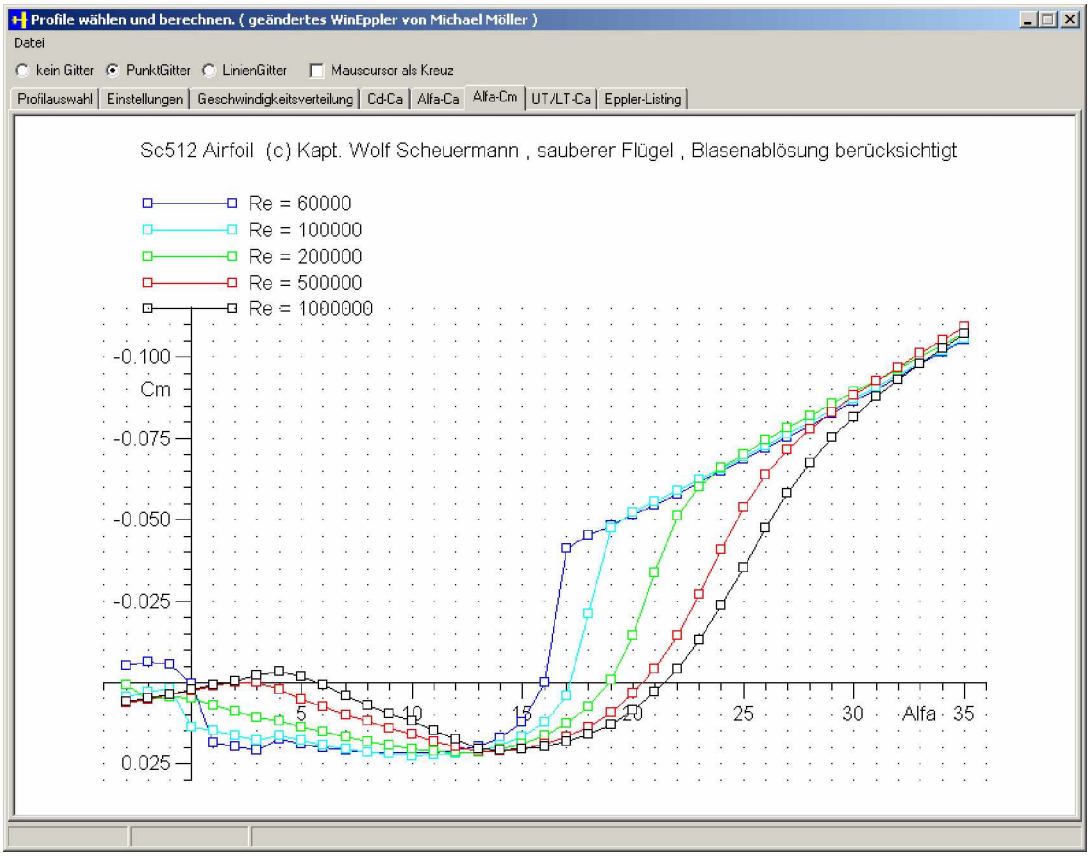
Daten des Flügelprofils Sc512

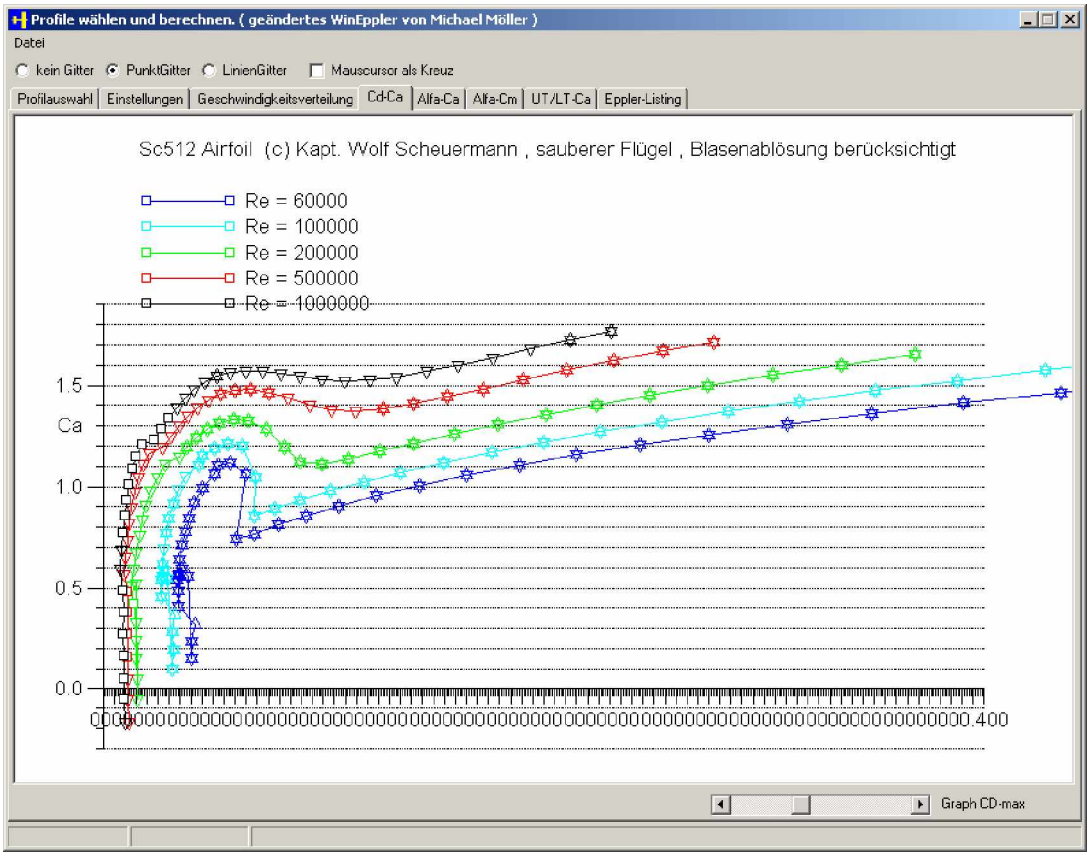


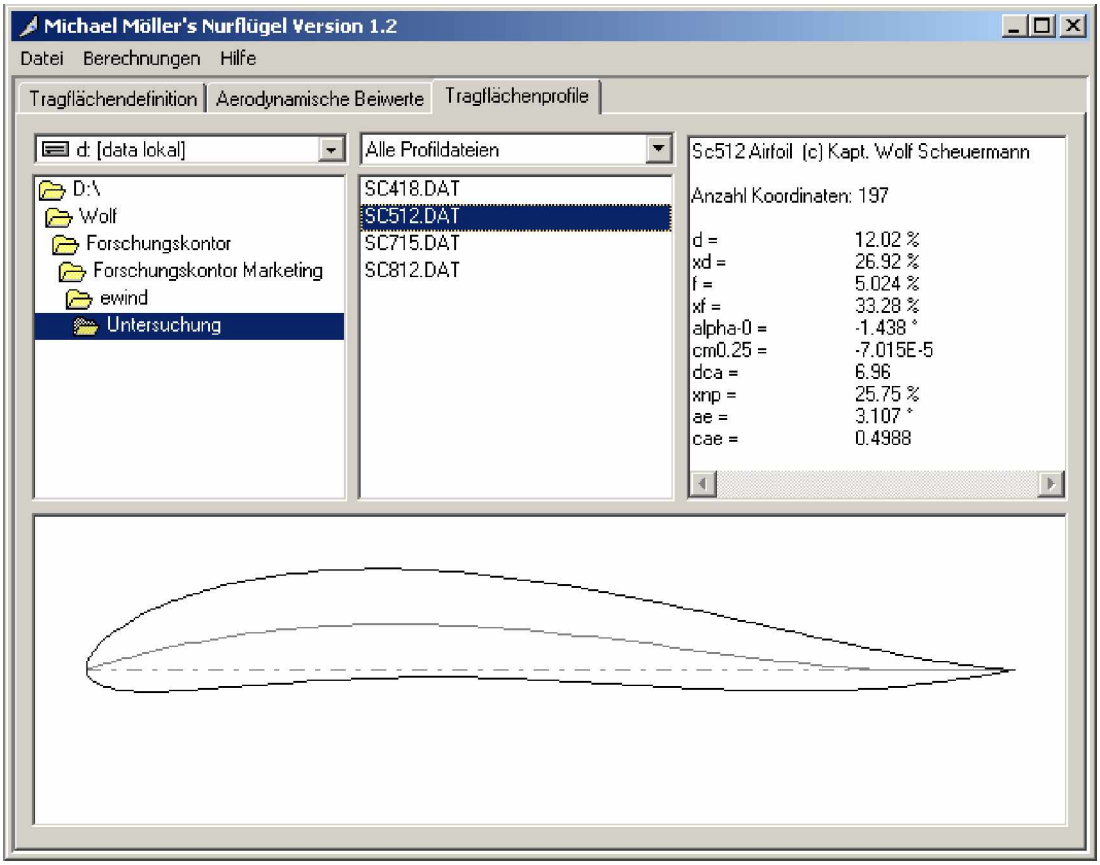
```
Sc512 Airfoil
01-21-2009 (c) Forschungskontor, Dipl.-Ing. Kapt.(AG) Wolf Scheuermann
Aerodynamic - Conformal Mapping - Complex Potential

z : circle centered on m1 with radius 1
intermediate v.Mises Airfoil:   z1 = z + b1/z + b2/z^2
Theodorsen Airfoil:   z2 = z1 + a1/z1 + a2/z1^2 + a3/z1^3

-0.10000 = Re(m1)
0.21160 = Im(m1)

-0.00600 = Re(b1)
-0.03880 = Im(b1)
0.00120 = Re(b2)
0.00000 = Im(b2)

0.75470 = Re(a1)
-0.22740 = Im(a1)
0.00010 = Re(a2)
0.09740 = Im(a2)
-0.00300 = Re(a3)
0.00000 = Im(a3)
```








Profile wählen und berechnen. (geändertes WinEppler von Michael Möller)
Datei
kein Gitter
PunktGitter
LinienGitter
Mauscursor als Kreuz
Profilauswahl
Einstellungen
Geschwindigkeitsverteilung
Cd-Ca
Alfa-Ca
Alfa-Cm
UT/LT-Ca
Eppler-Listing
Sc512 Airfoil (c) Kapt. Wolf Scheuermann , sauberer Flügel , Blasenablösung berücksichtigt
Re = 60000
Re = 100000
Re = 200000
Re = 500000
Re = 1000000
1.5
Ca
1.0
0.5
0.0
0.400
Graph CD-max


Michael Möller's Nurflügel Version 1.2
Datei
Berechnungen
Hilfe
Tragflächendefinition
Aerodynamische Beiwerte
Tragflächenprofile
d: [data lokal]
D:\
Wolf
Forschungskontor
Forschungskontor Marketing
ewind
Untersuchung
Alle Profildateien
SC418.DAT
SC512.DAT
SC715.DAT
SC812.DAT
Sc512 Airfoil (c) Kapt. Wolf Scheuermann
Anzahl Koordinaten: 197
d = 12.02 %
xd = 26.92 %
f = 5.024 %
xf = 33.28 %
alpha-0 = -1.438 °
cm0.25 = -7.015E-5
dca = 6.96
xnp = 25.75 %
ae = 3.107 °
cae = 0.4988